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SECTION 5

PERFORMANCE

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**SECTION 5
PERFORMANCE**

5.1 GENERAL

All of the required (FAA regulations) and complementary performance information applicable to the Archer II is provided by this section.

Performance information associated with those optional systems and equipment which require handbook supplements is provided by Section 9 (Supplements).

**5.3 INTRODUCTION TO PERFORMANCE AND FLIGHT
PLANNING**

The performance information presented in this section is based on measured Flight Test Data corrected to I.C.A.O. standard day conditions and analytically expanded for the various parameters of weight, altitude, temperature, etc.

The performance charts are unfactored and do not make any allowance for varying degrees of pilot proficiency or mechanical deterioration of the aircraft. This performance, however, can be duplicated by following the stated procedures in a properly maintained airplane.

Effects of conditions not considered on the charts must be evaluated by the pilot, such as the effect of soft or grass runway surface on takeoff and landing performance, or the effect of winds aloft on cruise and range performance. Endurance can be grossly affected by improper leaning procedures, and inflight fuel flow and quantity checks are recommended.

REMEMBER! To get chart performance, follow the chart procedures.

**SECTION 5
PERFORMANCE**

**PIPER AIRCRAFT CORPORATION
PA-28-181, ARCHER II**

The information provided by paragraph 5.5 (Flight Planning Example) outlines a detailed flight plan using the performance charts in this section. Each chart includes its own example to show how it is used.

WARNING

Performance information derived by extrapolation beyond the limits shown on the charts should not be used for flight planning purposes.

5.5 FLIGHT PLANNING EXAMPLE

(a) Aircraft Loading

The first step in planning the flight is to calculate the airplane weight and center of gravity by utilizing the information provided by Section 6 (Weight and Balance) of this handbook.

The basic empty weight for the airplane as licensed at the factory has been entered in Figure 6-5. If any alterations to the airplane have been made effecting weight and balance, reference to the aircraft logbook and Weight and Balance Record (Figure 6-7) should be made to determine the current basic empty weight of the airplane.

Make use of the Weight and Balance Loading Form (Figure 6-11) and the C.G. Range and Weight graph (Figure 6-15) to determine the total weight of the airplane and the center of gravity position.

After proper utilization of the information provided, the following weights have been determined for consideration in the flight planning example.

The landing weight cannot be determined until the weight of the fuel to be used has been established [refer to item (g)(1)].

(1) Empty Weight	1400 lbs.
(2) Occupants (2 x 170 lbs.)	340 lbs.
(3) Baggage and Cargo	360 lbs.
(4) Fuel (6 lb./gal. x 50)	300 lbs.
(5) Takeoff Weight	2400 lbs.
(6) Landing Weight	
(a)(5) minus (g)(1), (2400 lbs. minus 129 lbs.)	2271 lbs.

The takeoff weight is below the maximum of 2550 lbs. and the weight and balance calculations have determined that the C.G. position is within the approved limits.

(b) Takeoff and Landing

After determining the aircraft loading, all aspects of takeoff and landing must be considered.

Conditions of the departure and destination airport must be acquired, evaluated and maintained throughout the flight.

Apply the departure airport conditions and takeoff weight to the appropriate Takeoff Performance graph (Figure 5-7 or 5-9) to determine the length of runway necessary for the takeoff and/or the barrier distance.

The landing distance calculations are performed in the same manner using the existing conditions at the destination airport and, when established, the landing weight.

The conditions and calculations for the example flight are listed below. The takeoff and landing distances required for the example flight have fallen well below the available runway lengths.

	Departure Airport	Destination Airport
(1) Pressure Altitude	2000 ft.	2300 ft.
(2) Temperature	21°C	21°C
(3) Wind Component (Headwind)	10 KTS	5 KTS
(4) Runway Length Available	7000 ft.	4500 ft.
(5) Runway Required	950 ft.*	825 ft.**

NOTE

The remainder of the performance charts used in this flight plan example assume a no wind condition. The effect of winds aloft must be considered by the pilot when computing climb, cruise and descent performance.

*reference Figure 5-13

**reference Figure 5-37

(c) Climb

The next step in the flight plan is to determine the necessary climb segment components.

The desired cruise pressure altitude and corresponding cruise outside air temperature values are the first variables to be considered in determining the climb components from the Time, Distance and Fuel to Climb graph (Figure 5-17). After the time, distance and fuel for the cruise pressure altitude and outside air temperature values have been established, apply the existing conditions at the departure field to the graph (Figure 5-17). Now, subtract the values obtained from the graph for the field of departure conditions from those for the cruise pressure altitude.

The remaining values are the true fuel, distance and time components for the climb segment of the flight plan corrected for field pressure altitude and temperature.

The following values were determined from the above instructions in the flight planning example.

(1) Cruise Pressure Altitude	6000 ft.
(2) Cruise OAT	13°C
(3) Time to Climb (11.5 min. minus 3 min.)	8.5 min.*
(4) Distance to Climb (16 minus 4.5 naut. miles)	11.5 naut. miles*
(5) Fuel to Climb (2 gal. minus 1 gal.)	1 gal.*

(d) Descent

The descent data will be determined prior to the cruise data to provide the descent distance for establishing the total cruise distance.

Utilizing the cruise pressure altitude and OAT, determine the basic time, distance and fuel for descent (Figure 5-31). These figures must be adjusted for the field pressure altitude and temperature at the destination airport. To find the necessary adjustment values, use the existing pressure altitude and temperature conditions at the destination airport as variables to find the time, distance and fuel

*reference Figure 5-17

values from the graph (Figure 5-31). Now, subtract the values obtained from the field conditions from the values obtained from the cruise conditions to find the true time, distance and fuel values needed for the flight plan.

The values obtained by proper utilization of the graphs for the descent segment of the example are shown below.

- | | | |
|-----------------------------|--|-------------------|
| (1) Time to Descend | | |
| (16 min. minus 7.5 min.) | | 8.5 min.* |
| (2) Distance to Descend | | |
| (35 minus 14.5 naut. miles) | | 20.5 naut. miles* |
| (3) Fuel to Descend | | |
| (2 gal. minus 1 gal.) | | 1 gal.* |

(e) Cruise

Using the total distance to be traveled during the flight, subtract the previously calculated distance to climb and distance to descend to establish the total cruise distance. Refer to the appropriate Avco Lycoming Operator's Manual when selecting the cruise power setting. The established pressure altitude and temperature values and the selected cruise power should now be utilized to determine the true airspeed from the appropriate Speed Power graph (Figure 5-21 or 5-23).

Calculate the cruise fuel flow for the cruise power setting from the information provided by the Avco Lycoming Operator's Manual.

The cruise time is found by dividing the cruise distance by the cruise speed and the cruise fuel is found by multiplying the cruise fuel flow by the cruise time.

The cruise calculations established for the cruise segment of the flight planning example are as follows:

- | | | |
|-----------------------------------|--|-----------------|
| (1) Total Distance | | 314 naut. miles |
| (2) Cruise Distance | | |
| (e)(1) minus (c)(4) minus (d)(2), | | |
| (314 minus 11.5 minus 20.5) | | 282 naut. miles |

*reference Figure 5-31

(3) Cruise Power	65% rated power
(4) Cruise Speed	110 KTS TAS*
(5) Cruise Fuel Consumption	7.6 GPH
(6) Cruise Time	
(e)(2) divided by (e)(4), (282 naut. miles divided by 110 KTS)	2.56 hrs.
(7) Cruise Fuel	
(e)(5) multiplied by (e)(6), (7.6 GPH multiplied by 2.56 hrs.)	19.5 gal.

(f) Total Flight Time

The total flight time is determined by adding the time to climb, the time to descend and the cruise time. Remember! The time values taken from the climb and descent graphs are in minutes and must be converted to hours before adding them to the cruise time.

The following flight time is required for the flight planning example.

(1) Total Flight Time	
(c)(3) plus (d)(1) plus (e)(6), (.14 hrs. plus .14 hrs. plus 2.56 hrs.)	2.84 hrs.

(g) Total Fuel Required

Determine the total fuel required by adding the fuel to climb, the fuel to descend and the cruise fuel. When the total fuel (in gallons) is determined, multiply this value by 6 lb./gal. to determine the total fuel weight used for the flight.

The total fuel calculations for the example flight plan are shown below.

(1) Total Fuel Required	
(c)(5) plus (d)(3) plus (e)(7), (1 gal. plus 1 gal. plus 19.5 gal.)	21.5 gal.
(21.5 gal. multiplied by 6 lb./gal.)	129 lbs.

*reference Figure 5-23

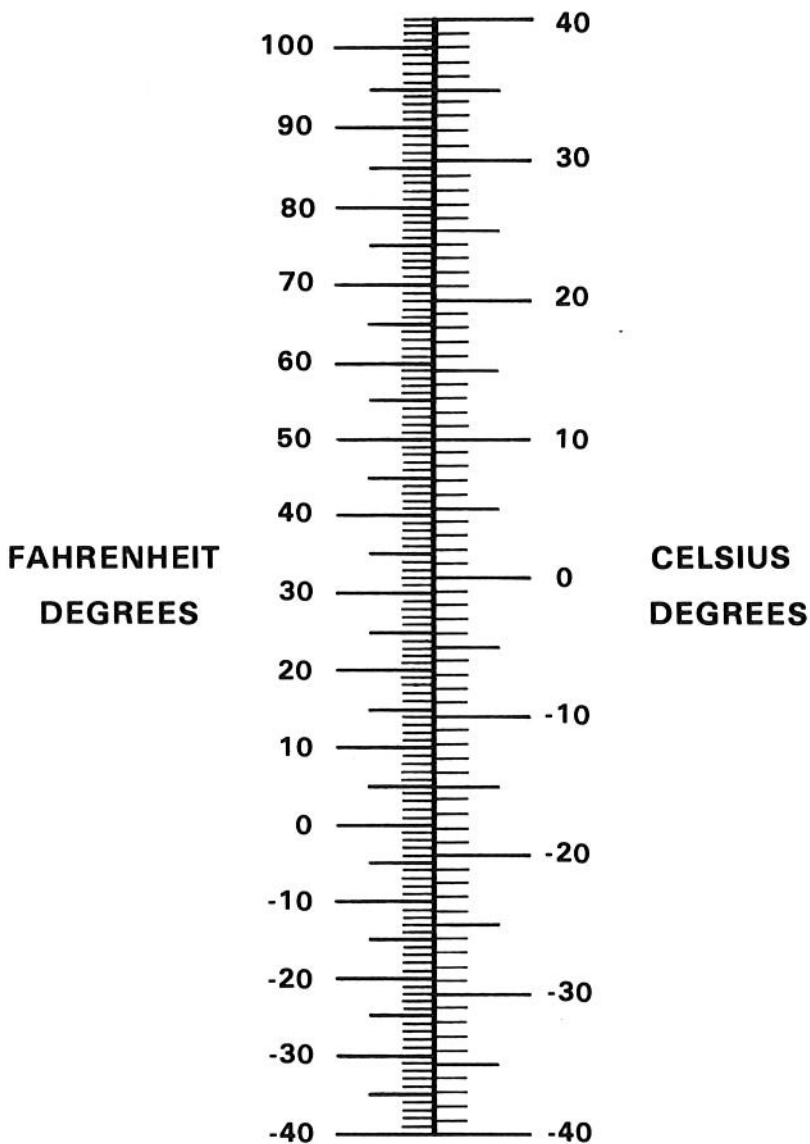
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5.7 PERFORMANCE GRAPHS

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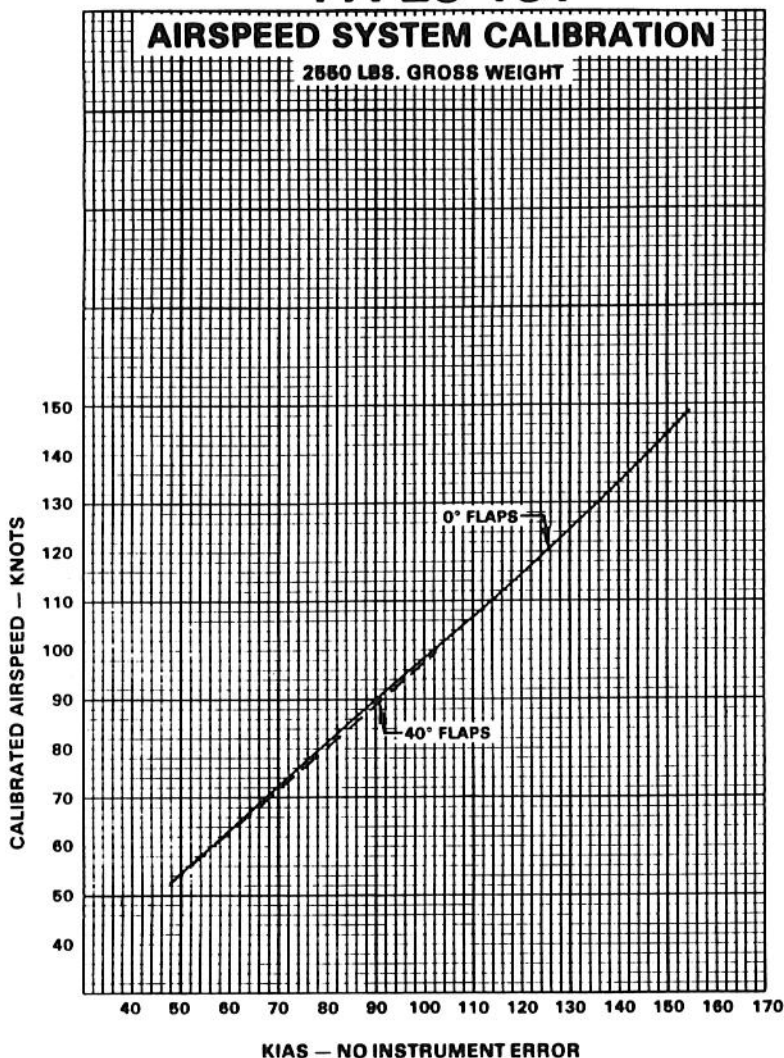
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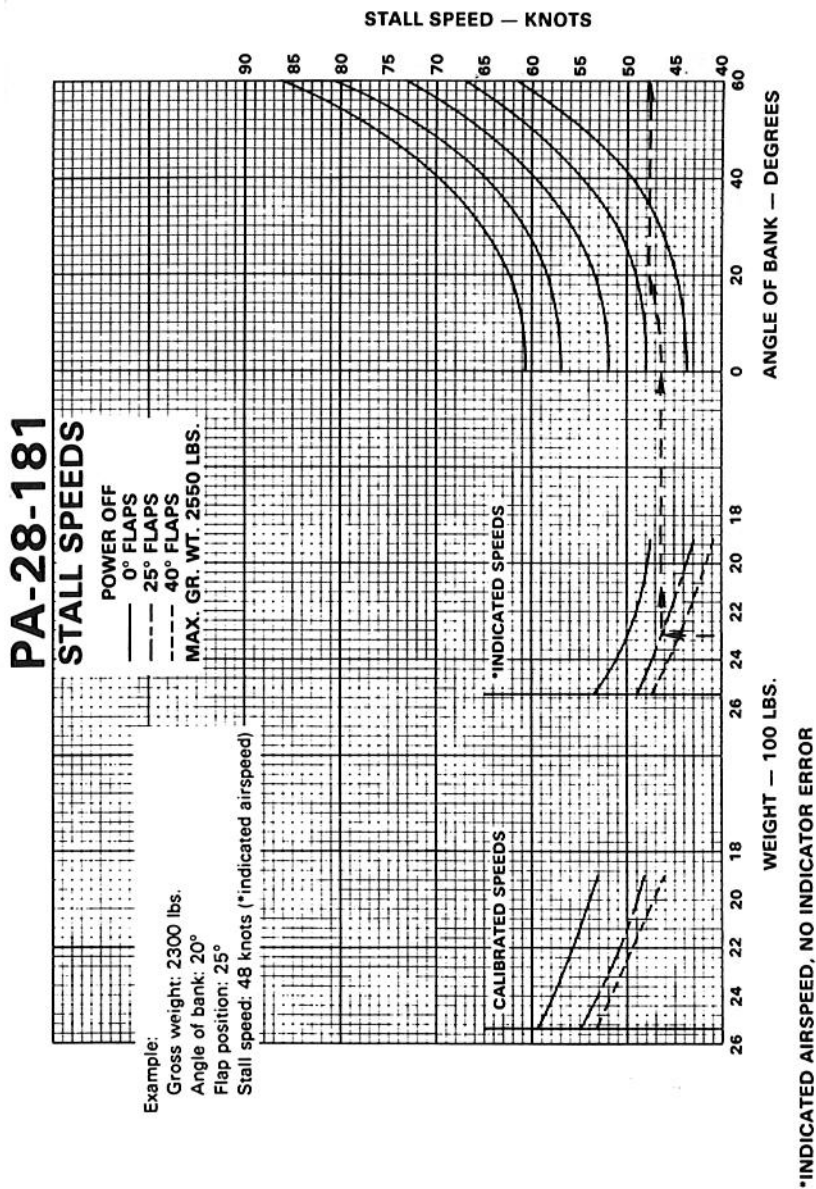
TEMPERATURE CONVERSION

Figure 5-1

PA-28-181

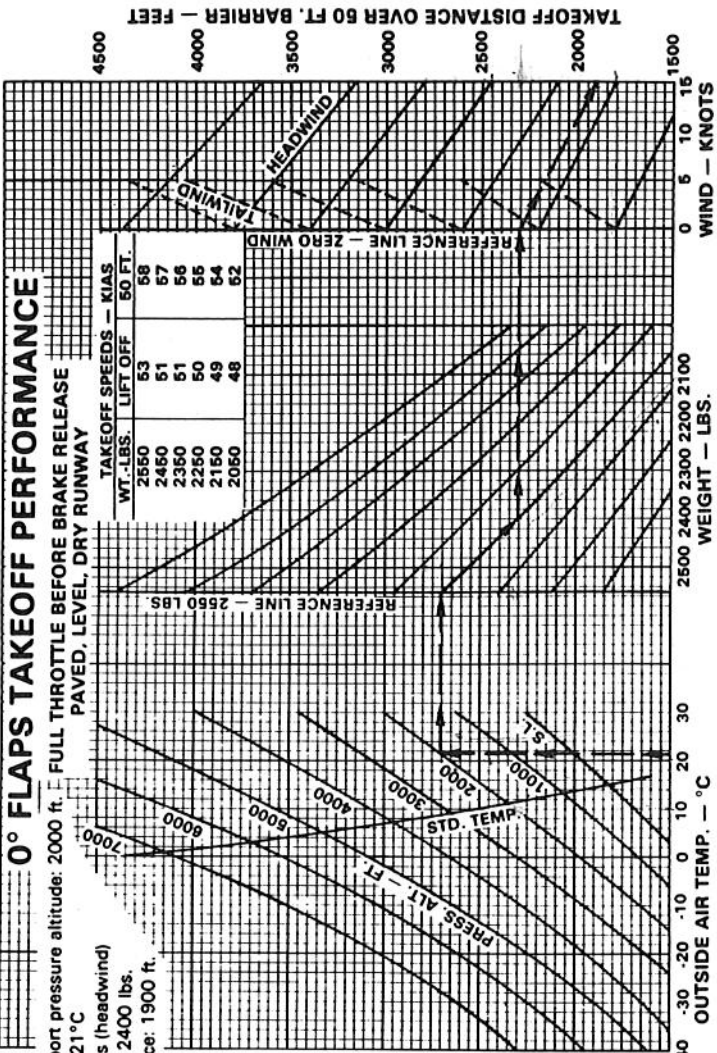


AIRSPED SYSTEM CALIBRATION
Figure 5-3



STALL SPEEDS
Figure 5-5

PA-28-181



FLAPS UP TAKEOFF PERFORMANCE

Figure 5-7

PA-28-181

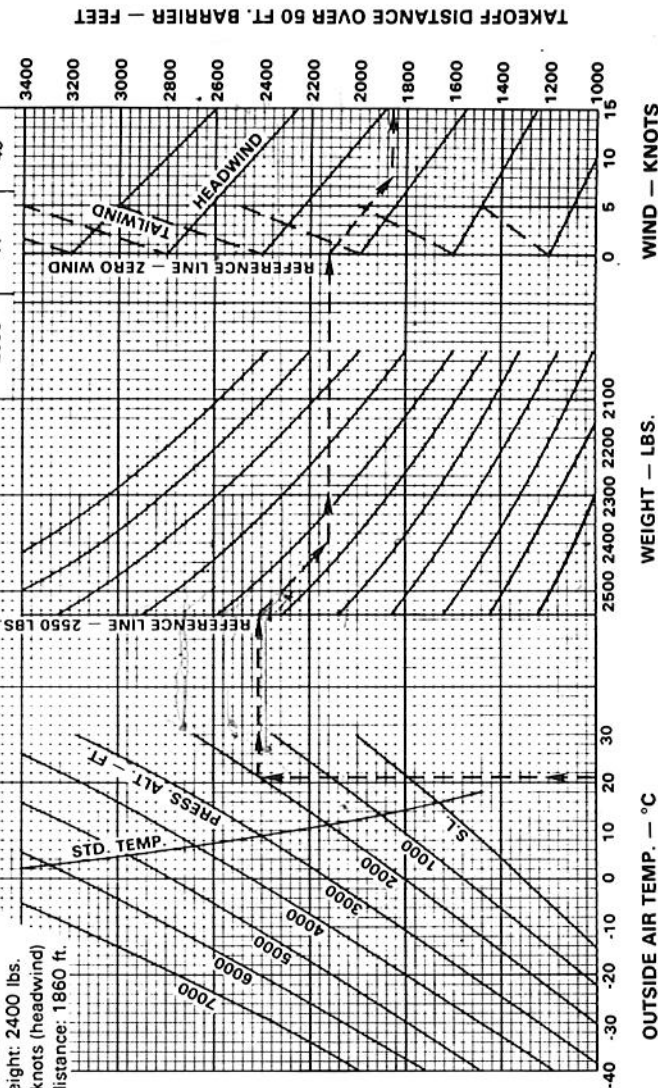
25° FLAPS TAKEOFF PERFORMANCE

FULL THROTTLE BEFORE BRAKE RELEASE
PAVED, LEVEL, DRY RUNWAY

TAKEOFF SPEEDS — KIAS	
WT. — LBS.	LIFT OFF — 50 FT.
2550	49
2450	47
2350	45
2250	43
2150	42
2050	41

Example:

Departure airport pressure altitude: 2000 ft.
Temperature: 21°C
Gross weight: 2400 lbs.
Wind: 8 knots (headwind)
Takeoff distance: 1860 ft.



25° FLAPS TAKEOFF PERFORMANCE

Figure 5-9

**SECTION 5
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PA-28-181

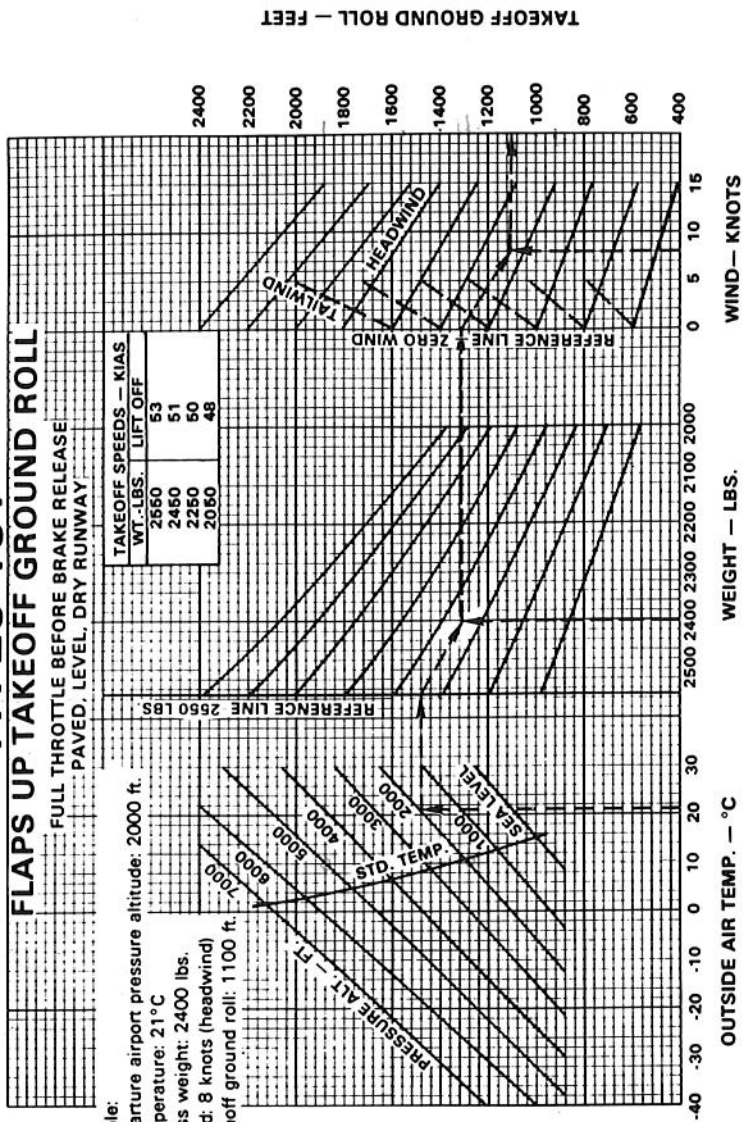
FLAPS UP TAKEOFF GROUND ROLL

FULL THROTTLE BEFORE BRAKE RELEASE
PAVED, LEVEL, DRY RUNWAY

Example:

Departure airport pressure altitude: 2000 ft.
Temperature: 21°C
Gross weight: 2400 lbs.
Wind: 8 knots (headwind)
Takeoff ground roll: 1100 ft.

TAKEOFF SPEEDS — KIAS	
WT. — LBS.	LIFT OFF
2650	53
2450	51
2250	50
2050	48



FLAPS UP TAKEOFF GROUND ROLL

Figure 5-11

PA-28-181

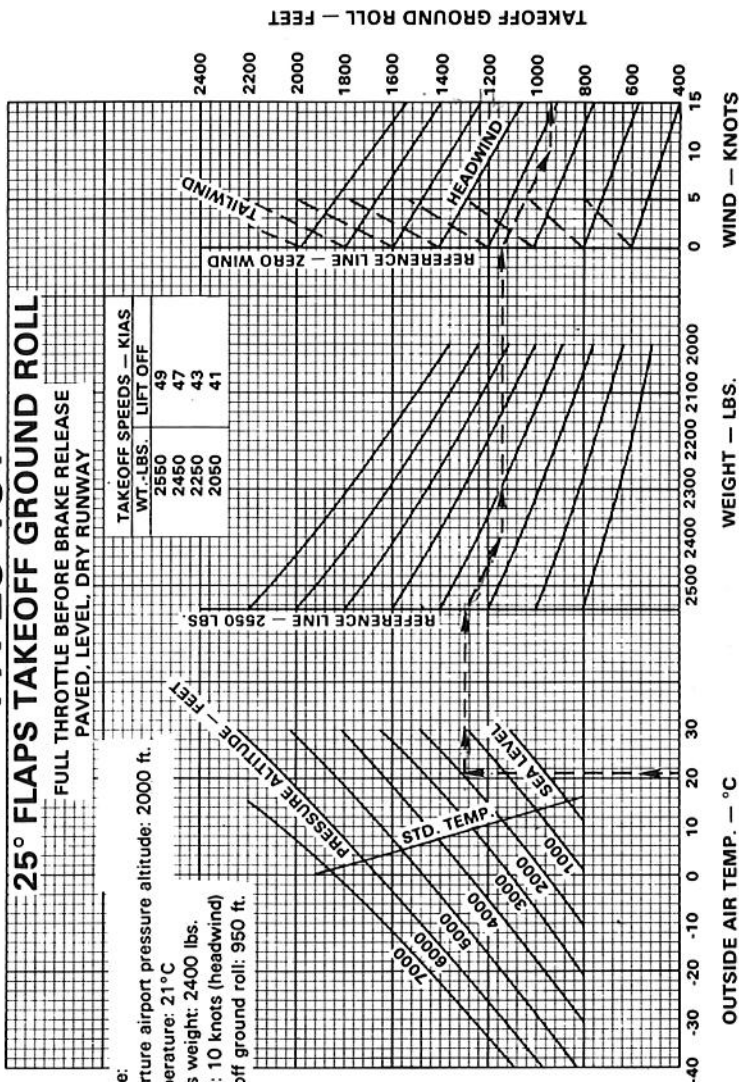
25° FLAPS TAKEOFF GROUND ROLL

FULL THROTTLE BEFORE BRAKE RELEASE
PAVED, LEVEL, DRY RUNWAY

Example:

Departure airport pressure altitude: 2000 ft.
Temperature: 21°C
Gross weight: 2400 lbs.
Wind: 10 knots (headwind)
Takeoff ground roll: 950 ft.

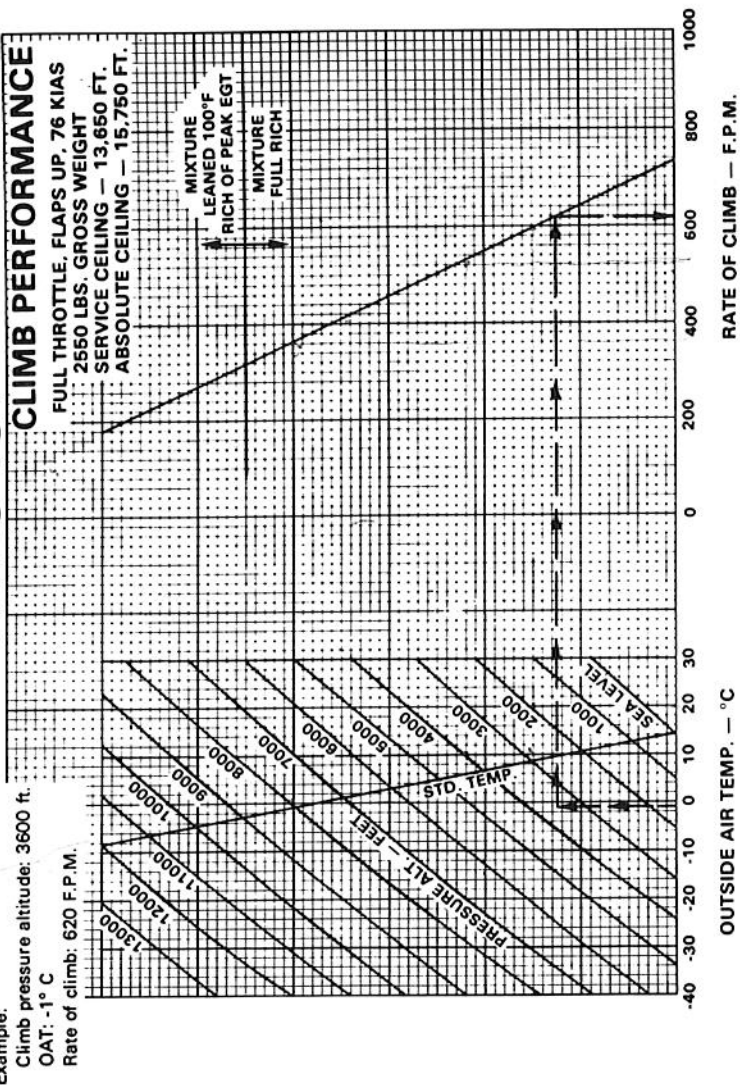
TAKEOFF SPEEDS — KIAS	
WT. — LBS.	LIFT OFF
2550	49
2450	47
2250	43
2050	41



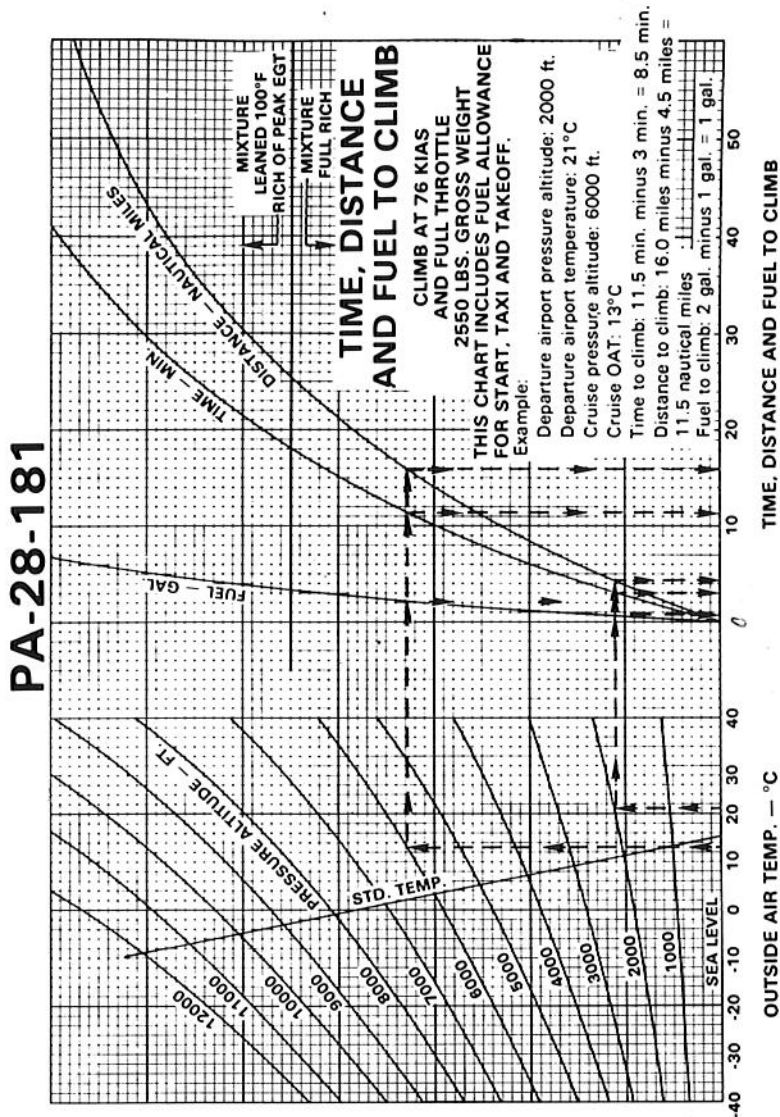
25° FLAPS TAKEOFF GROUND ROLL

Figure 5-13

PA-28-181

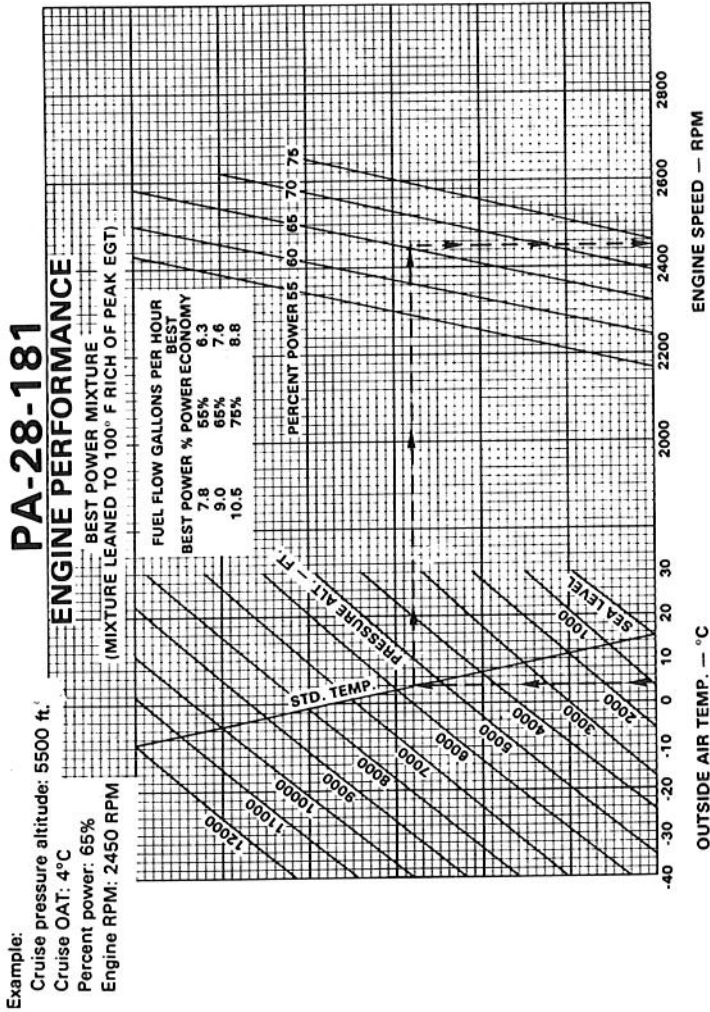


CLIMB PERFORMANCE
Figure 5-15



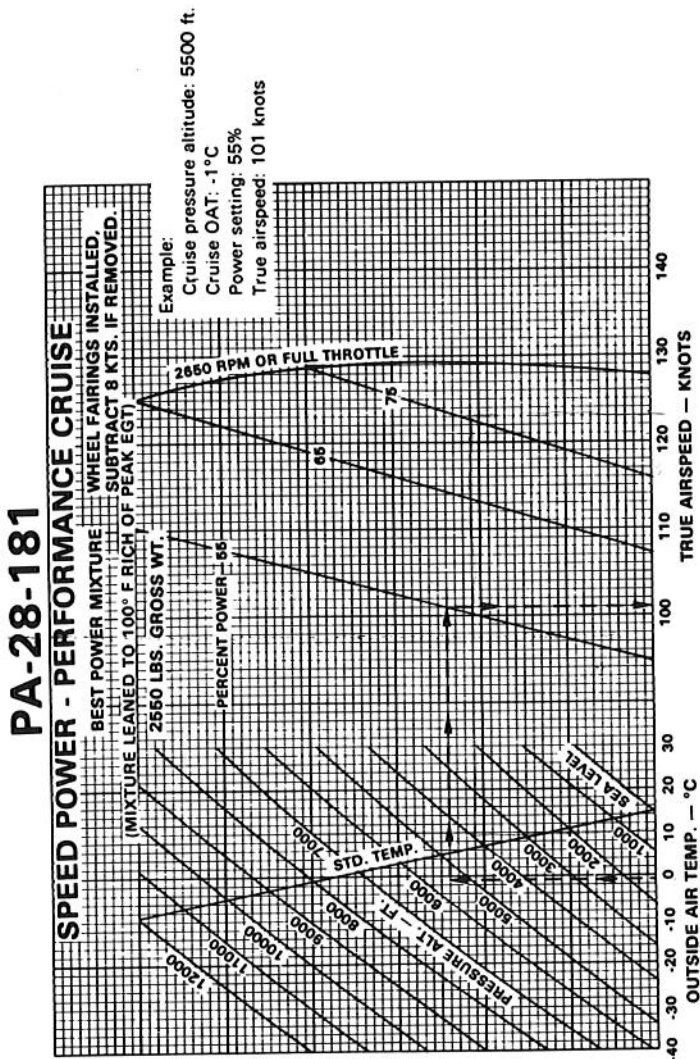
TIME, DISTANCE AND FUEL TO CLIMB

Figure 5-17



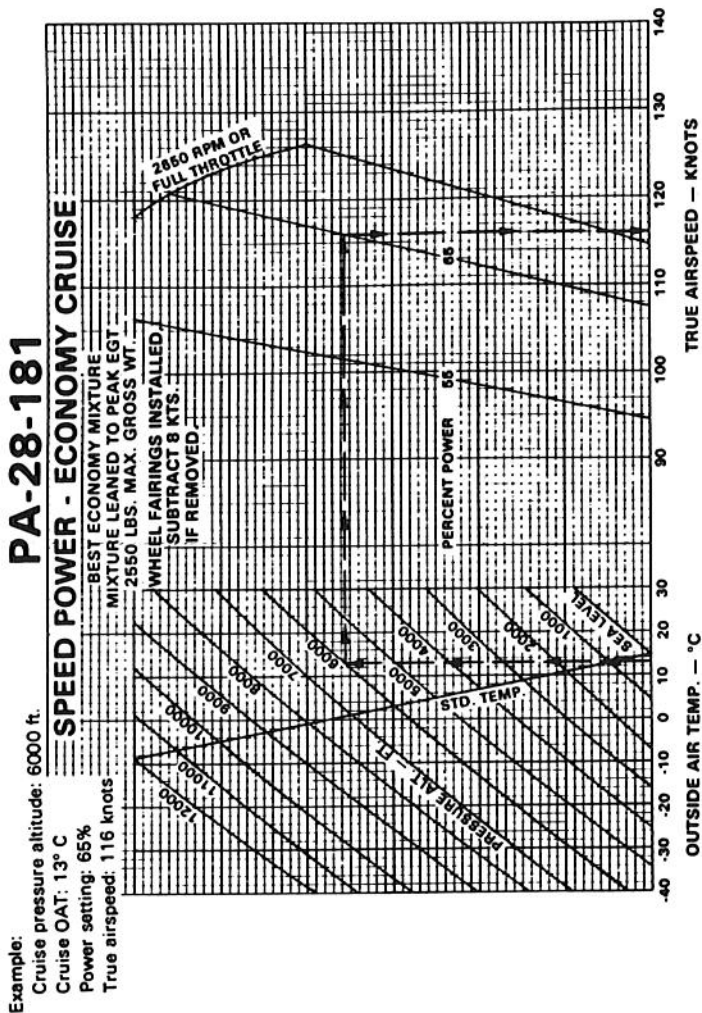
ENGINE PERFORMANCE

Figure 5-19



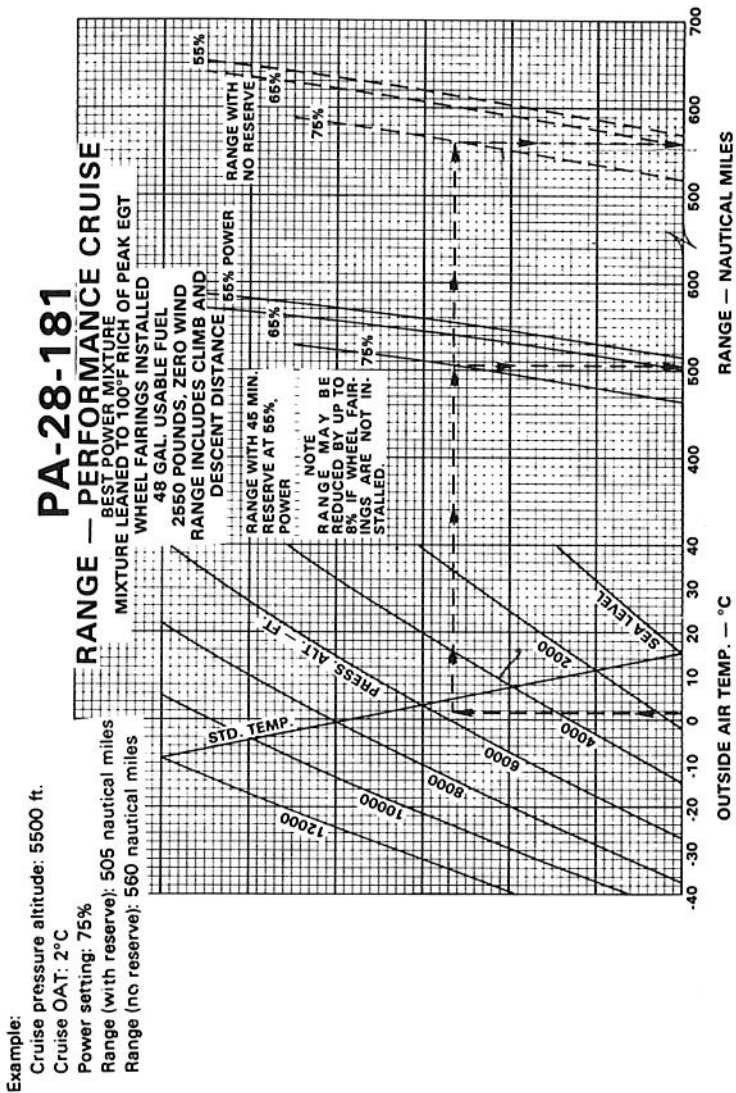
SPEED POWER - PERFORMANCE CRUISE

Figure 5-21



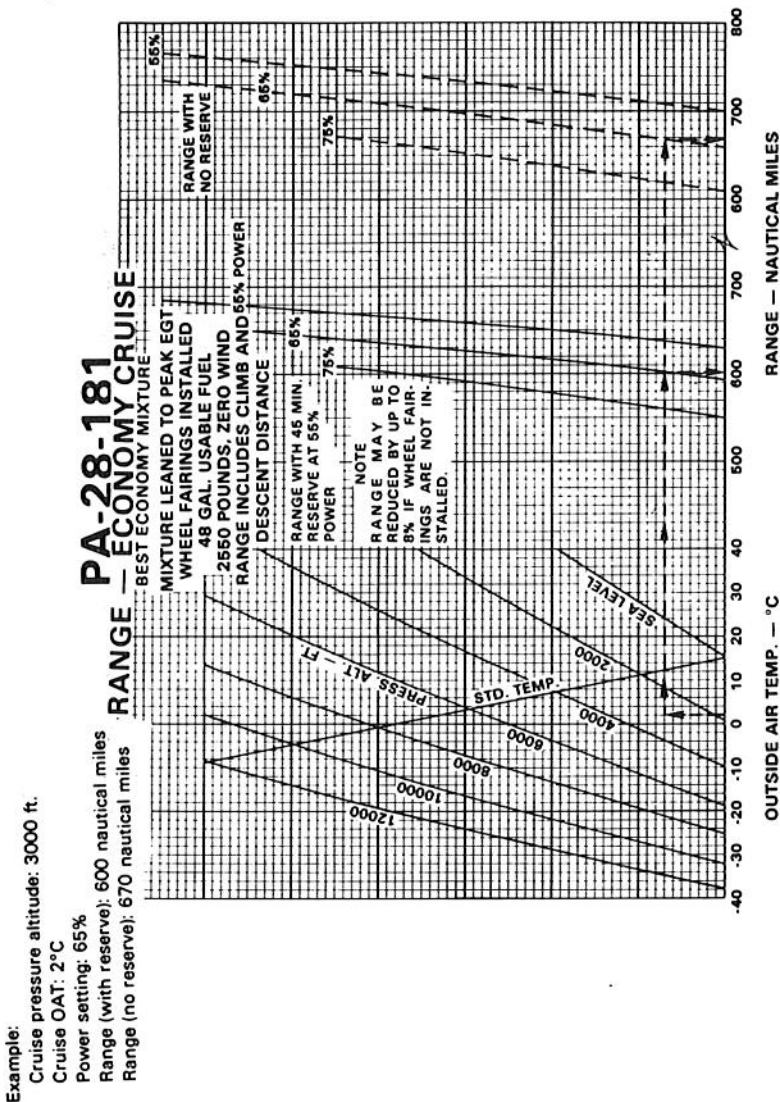
SPEED POWER - ECONOMY CRUISE

Figure 5-23



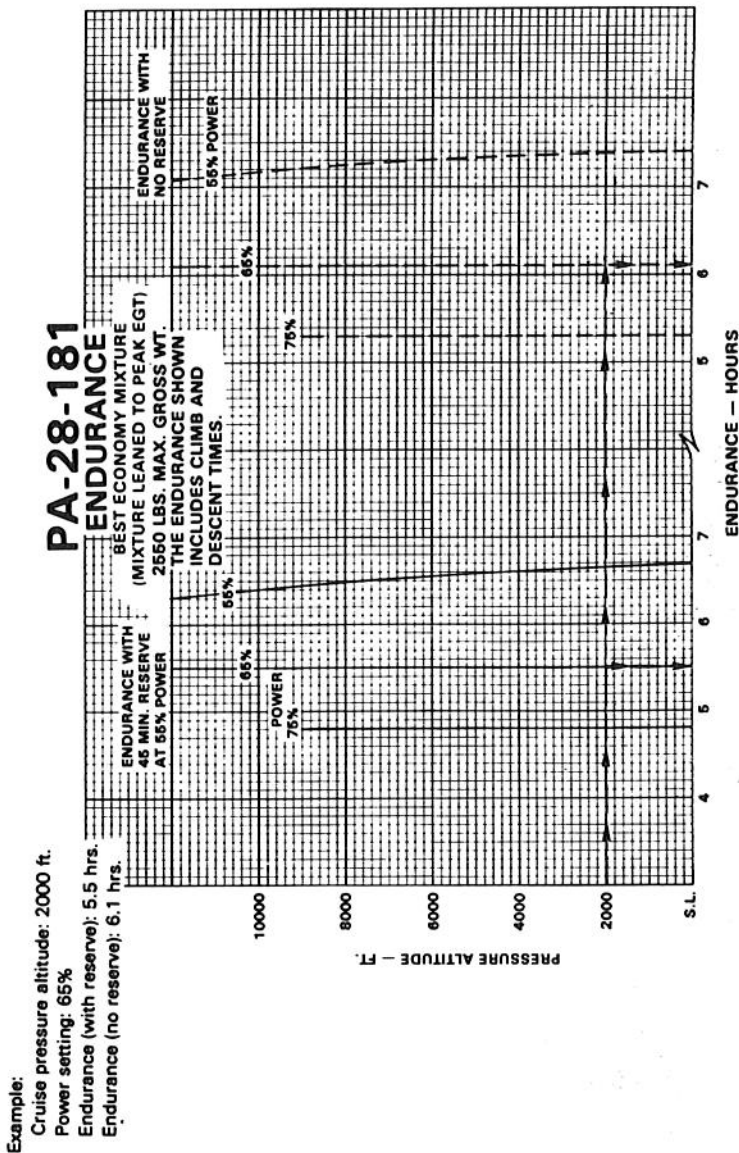
BEST POWER MIXTURE RANGE

Figure 5-25



BEST ECONOMY MIXTURE RANGE

Figure 5-27



ENDURANCE
Figure 5-29

PA-28-181

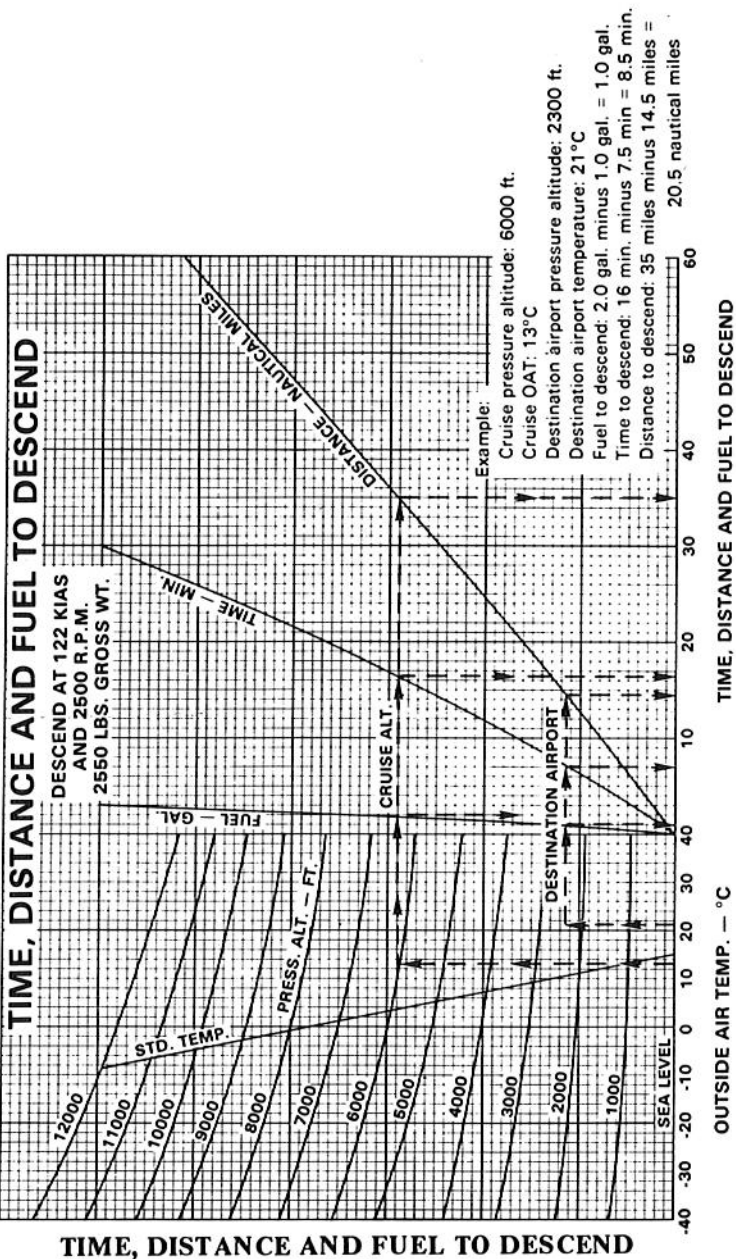
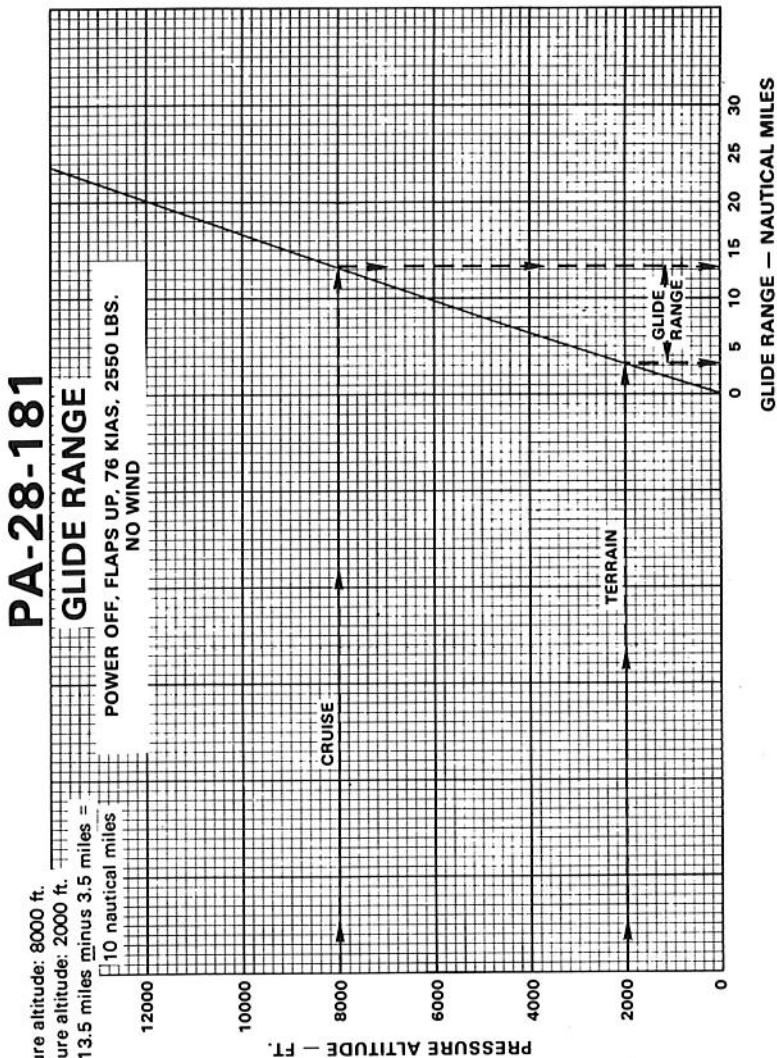


Figure 5-31

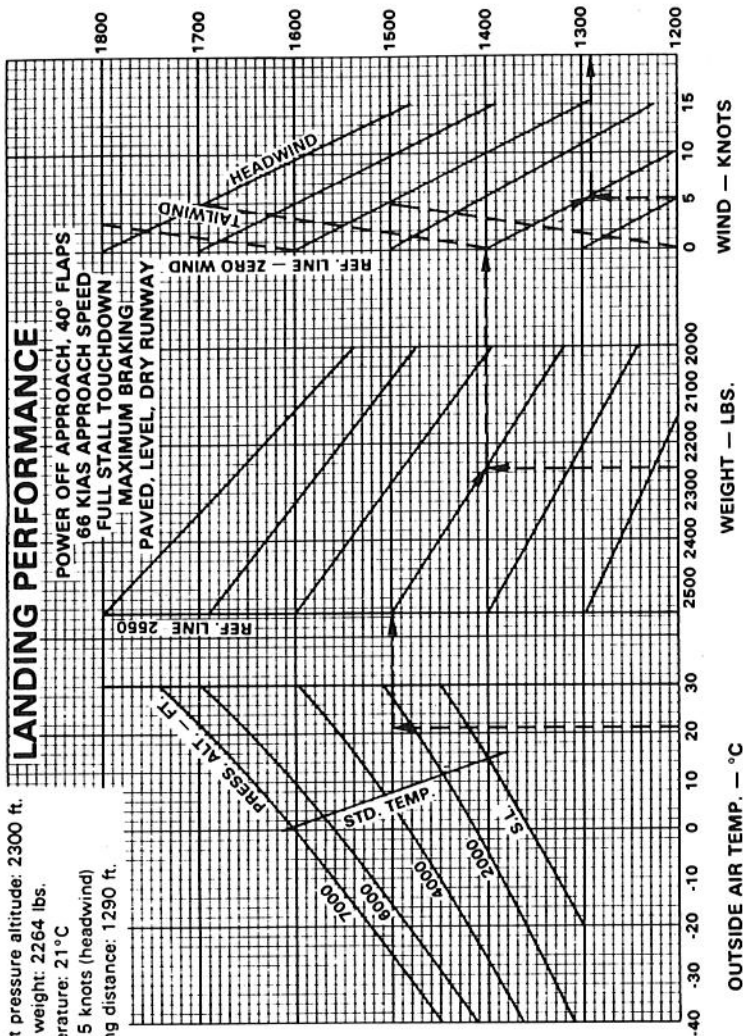


GLIDE RANGE
Figure 5-33

PA-28-181

LANDING PERFORMANCE

Example:
 Airport pressure altitude: 2300 ft.
 Gross weight: 2264 lbs.
 Temperature: 21°C
 Wind: 5 knots (headwind)
 Landing distance: 1290 ft.



LANDING PERFORMANCE

Figure 5-35

